

Calculate the ideal chamber pressure for a rocket motor having a K_n of 180

$$K_n = \frac{A_B \text{ (burn area)}}{A_* \text{ (throat area)}} = 180$$

Steady state chamber pressure = P_0

$$P_0 = \left[\frac{A_B}{A_*} \frac{a P_p}{\sqrt{\frac{\kappa}{RT_0} \left(\frac{2}{\kappa+1}\right)^{\frac{\kappa+1}{\kappa-1}}}} \right]^{\frac{1}{1-n}}$$

$$P_0 = \left[180 \cdot \frac{a P_p}{\sqrt{\frac{\kappa}{RT_0} \left(\frac{2}{\kappa+1}\right)^{\frac{\kappa+1}{\kappa-1}}}} \right]^{\frac{1}{1-n}}$$

for KNSU, $\kappa = 1.33$

$$T_0 = 1720 \text{ K}$$

$$M = 41.98 \text{ kg/kmol}$$

$$R' = 8314 \text{ J-m/kmol-K} \quad \text{universal gas constant}$$

$$R = \frac{R'}{M} \Rightarrow R = 198 \text{ Nm/kg-K}$$

wrong
at } by St Robert's Law, burn rate can be expressed in terms of chamber pressure

$$r = a P_0^n \quad \begin{cases} a: \text{burn rate coefficient} \\ n: \text{constant} \end{cases}$$

by chemistry of KNSU:

$$a = 0.0665 \text{ in/sec-psi}^n \quad P_p$$

$$n = 0.319$$

$$a = 0.0665 \frac{\text{in}}{\text{sec-psi}^n} \cdot \frac{1 \text{ m}}{39.37 \text{ in}} \cdot \frac{1 \text{ sec-psi}}{(6895)^{0.319} \text{ Pa}^n} = 0.000$$

Must convert to SI units

plug in, get:

$$P_0 = 4029000 \text{ Pa} = 4.03 \text{ MPa} = 584 \text{ psi}$$

Calculating ideal chamber pressure for APCP
 Propellant with a k_n of 180-250
 (assume 215)

$$k = 1.21 \text{ (ratio of specific heats)}$$

$$T_0 = 3500 \text{ K (adiabatic flame temp)}$$

$$M = 23.67 \text{ g/mol}$$

$$R' = 8314 \text{ N-m/kgmol-K}$$

$$R = \frac{R'}{M} = \frac{8314}{23.67} = 351 \text{ N-m/kg-K}$$

$$\rho_p = 1680 \text{ g/m}^3 \text{ (propellant density)}$$

$$a = 3.5170541 \text{ (burn rate coefficient)}$$

$$n = 0.3273 \text{ (burn rate exponent)}$$

$$P_0 = \left[\frac{A_b}{A^*} \frac{a \rho_p}{\sqrt{\frac{k}{RT_0} \left(\frac{2}{k+1} \right)^{\frac{k+1}{k-1}}}} \right]^{\frac{1}{1-n}}$$

Calc one component at a time

$$\frac{A_b}{A^*} = k_n = 215$$

$$\frac{k}{RT_0} = 1.06773269 \text{ E-6}$$

$$\frac{2}{k+1} = 0.9049773756$$

$$\frac{k+1}{k-1} = 10.52380952$$

$$\frac{1}{1-n} = 1.486546752$$

$$a \rho_p = 0.0590865089$$

$$P_0 = \left[215 \cdot \frac{0.0590865089}{\sqrt{1.06 \text{ E-6} (0.90)^{10.52}}} \right]^{1.486}$$

$$P_0 = 2635775.616 \text{ Pa}$$

$$P_0 = 382 \text{ psi} = 26.2 \text{ atm}$$

Calculating optimum expansion ratio

$$\frac{A^*}{A_e} = \left(\frac{k+1}{2}\right)^{\frac{1}{k-1}} \left(\frac{P_e}{P_0}\right)^{\frac{1}{k}} \sqrt{\left(\frac{k+1}{k-1}\right) \left[1 - \left(\frac{P_e}{P_0}\right)^{\frac{k-1}{k}}\right]}$$

Solve everything involving k

$$\frac{k+1}{2} = 1.105$$

$$\frac{1}{k-1} = 4.7619$$

$$\frac{1}{k} = 0.826$$

$$\frac{k+1}{k-1} = 10.5238$$

$$\frac{k-1}{k} = 0.1735$$

$$\frac{P_e}{P_0} = \frac{1}{26} = 0.0384$$

$$\frac{A^*}{A_e} = (1.105)^{4.7619} (0.0384)^{0.826} \sqrt{(10.52)(1 - (0.0384)^{0.1735})}$$

$$\frac{A^*}{A_e} = 0.232201$$

$$\frac{A_e}{A^*} = 4.30$$

optimal expansion ratio